

SERVICE BULLETIN No. L13/004

Sheet 1 of 2

Subject: Rear fuselage section of the L13 a/c

Reason: Cracking of the rear fuselage skin panel within the joint gap between stabilizer hinge and frame No. 14, due to fatigue.

Description: Stiffening of the rear fuselage section by rivetting the fin skin panel to prevent its cracking due to fatigue.

Accomplishment instructions:

1. Mark and drill two holes dia 2.7mm in the fin skin and rear fuselage on either side to Fig.1, det.view A (shortest length of the rivet centres from the edge is 5mm)
2. Install the rivets into the drilled holes with their heads from outside the fuselage and attach the skin panel by rivetting.

Material information

Qty	Title	Dwg.No.
4	Rivet	2.6x6 CSN 02 2304.5

Compliance: The modification is to be incorporated on all the aircraft 1st series through 15 a/c 12th series.

Material: Material required will be supplied by the manufacturer -LET n.p. Uh. Hradiste, Czechoslovakia, who will also cover any costs incurred.

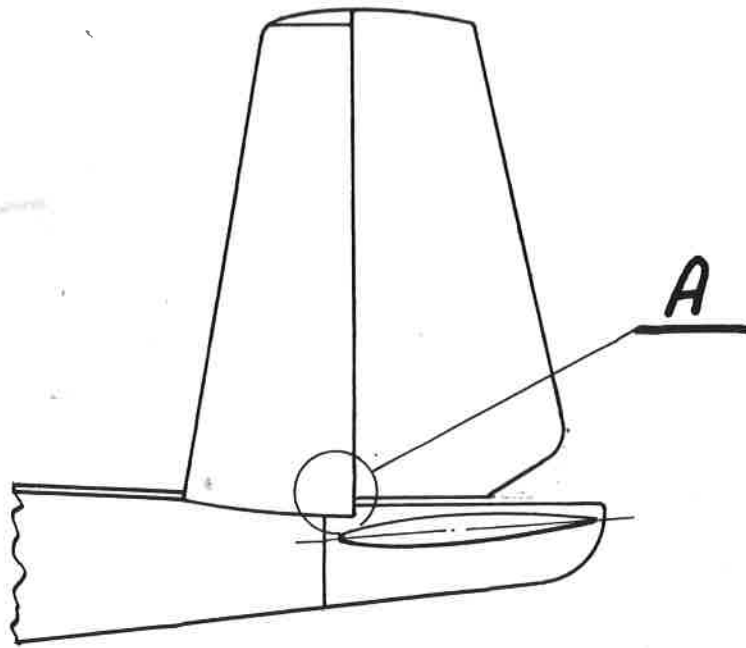
Manufacturer's
representative

Kosek

Customer's
representative

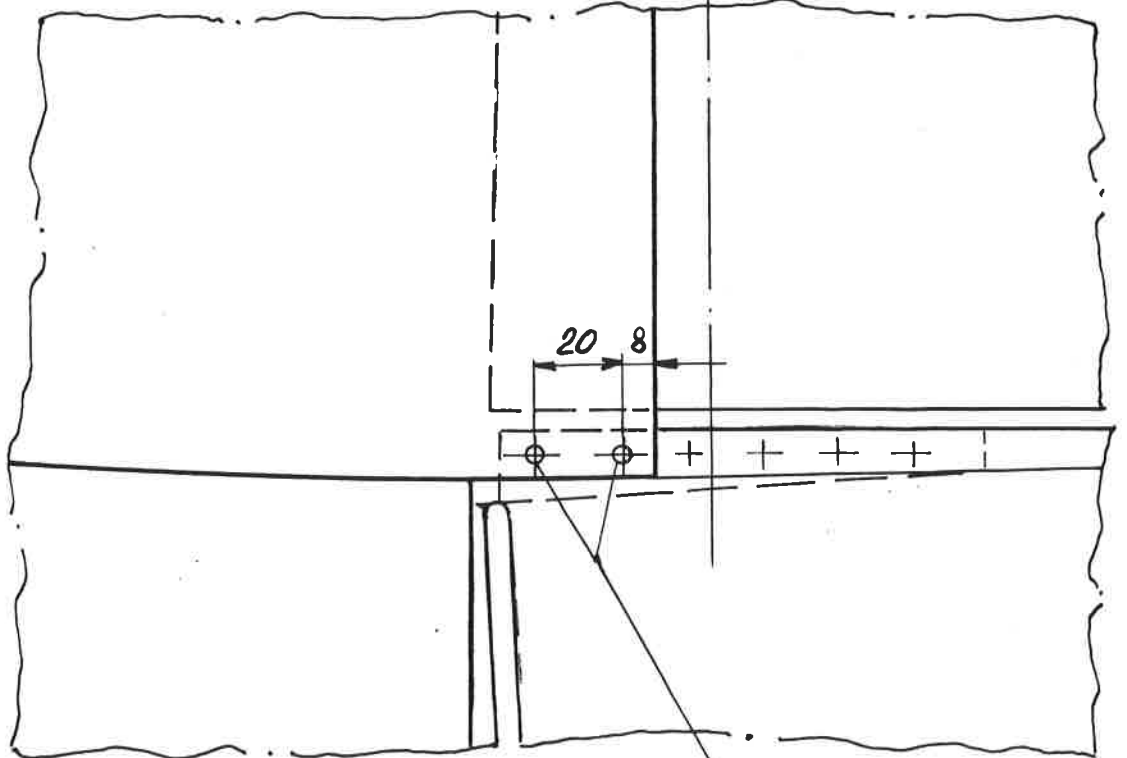
Kopřiva

Obn. 1



Det. A

Osa otáčení směr. korm.



5DuV 2,6x6 P
4 ks. na stroj.

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L 13/N2 B